

SHUTTLE CRITICAL ITEMS LIST - ORBITER

SUBSYSTEM : EPD&C - COMM. & TRACK. FMEA NO 05-6PD-22702 -1 REV: 11/06/87

ASSEMBLY : PNL 014 & 015 CRIT. FUNC: 1R
 P/N RI : MC454-0026-2030 CRIT. HDW: 2
 P/N VENDOR: VEHICLE 102 103 104
 QUANTITY : 2 EFFECTIVITY: X X X
 : TWO, 1 FOR EA ALTIMETER PHASE(S): PL X LO X OO DO X LS

REDUNDANCY SCREEN: A-PASS B-PASS C-PASS
 PREPARED BY: APPROVED BY: APPROVED BY (NASA):
 DES 11/10/87 *[Signature]* C ELDER DES *[Signature]* 11/13/87 SSM *[Signature]*
 REL 11/10/87 *[Signature]* A L MASAI REL *[Signature]* 11-14-87 REL *[Signature]*
 QE 11/11/87 *[Signature]* J T COURSEN QE *[Signature]* 11/20/87
 REL EPDC *[Signature]*
 SSM EPDC *[Signature]*

ITEM:
 CIRCUIT BREAKER, RADAR ALTIMETER.

FUNCTION:
 PROVIDES CIRCUIT PROTECTION FOR DC BUSES MNA AND MNB. 2/CIRCUIT
 BREAKER, 3 AMP, 33V73A14CH24, 33V73A15CB23.

FAILURE MODE:
 FAILS OPEN, FAILS TO CONDUCT, FAILS TO CLOSE.

CAUSE(S):
 STRUCTURAL FAILURE, CONTAMINATION, VIBRATION, MECHANICAL SHOCK
 PROCESSING ANOMALY, THERMAL STRESS.

EFFECT(S) ON:
 (A) SUBSYSTEM (B) INTERFACES (C) MISSION (D) CREW/VEHICLE
 (A) NO EFFECT ON EPDC.
 (B) LOSS OF ALTITUDE DATA IF THAT ALTIMETER IS SELECTED.
 (C) NO EFFECT.
 (D) POSSIBLE LOSS OF CREW/VEHICLE AFTER TWO CIRCUIT BREAKER FAILURE
 SINCE PRECISE ALTITUDE DATA FOR CREW DETERMINATION OF SINK RATE IS
 REQUIRED FOR SAFE NIGHT LANDING OR LANDINGS ON RUNWAYS WITHOUT MSELs AN
 TO PREVENT POSSIBLE VEHICLE DAMAGE.

DISPOSITION & RATIONALE:
 (A) DESIGN (B) TEST (C) INSPECTION (D) FAILURE HISTORY (E) OPERATIONAL USE
 (A, B, C, D) REFER TO APPENDIX D, ITEM # 1, CIRCUIT BREAKER

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(B) TEST:

GROUND TURNAROUND TEST- VERIFY RADAR ALTIMETER POWER ON FROM THE D & C PANEL.

(E) OPERATIONAL USE

RADAR ALTIMETER DATA IS AVAILABLE AT 5,000 FEET. DE-ORBIT IS NOT ATTEMPTED IF CEILING IS LESS THAN 8,000 FEET (10,000 FEET IF NO MSBLS AVAILABLE) TO ENSURE GOOD VISIBILITY AT LOW ALTITUDE. MOST ORBITER RUNWAYS ARE EQUIPPED WITH MSBLS GROUND STATIONS WHICH PROVIDE A REDUNDANT SOURCE OF LOW ALTITUDE DATA DOWN TO 50 FEET. RADAR ALTIMETER DATA IS DISPLAYED ON THE HUD AND AVVI NEXT TO NAVIGATION ALTITUDE DATA. CREW CAN ISOLATE A FAILED RADAR ALTIMETER AND THEN SELECT THE OTHER ALTIMETER, IF AVAILABLE, OR DISREGARD RADAR ALTIMETER DATA AND RELY ON NAVIGATION ALTITUDE AND/OR VISUAL CUES.