

April 19, 1996

CRITICAL ITEMS LIST

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1) CIL ITEM : F200-BZ-01
 2) FMEA CODE : F200-BZ
 3) COMPONENT : VIBRATION SIGNAL PROCESSOR ELECTRONICS (VSPE) NPFTP AND NPQTP
 4) PART NUMBER : RET493
 5) SYSTEM/SUBSYSTEM : CONTROLLER/F200-KX (FASDDS REDLINE ACTIVE)
 6) FAILURE MODE : FAILURE OF VSPE

7) PREPARED : SSME RELIABILITY
 8) APPROVED :
 9) DATE : 04-19-96
 10) REVISION/CHANGE : -001/D
 11) EFFECTIVITY : -05
 12) HAZARD REFERENCE : SEE LISTINGS BELOW
 13) ECBD # : me3-01-3285

PHASE	FAILURE DESCRIPTION/EFFECT	CRITICALITY
SM 4-2	<p>FAILURE OF TWO CHANNELS OUTSIDE OF QUALIFICATION LIMITS CAUSES CHANNEL DISQUALIFICATION RESULTING IN A LOSS OF REDLINE PROTECTION. LOSS OF VEHICLE DUE TO TURBOPUMP FAILURE MAY RESULT IF EXCESSIVE VIBRATION OCCURS AND IS NOT DETECTED.</p> <p>REDUNDANCY SCREENS: CONTROLLER SYSTEM - TURBOPUMP SYSTEM: UNLIKE REDUNDANCY</p> <p>A: PASS. REDUNDANT HARDWARE ITEMS ARE CAPABLE OF CHECKOUT DURING NORMAL GROUND TURNAROUND. B: PASS. LOSS OF A REDUNDANT HARDWARE ITEM IS DETECTABLE DURING FLIGHT. C: PASS. LOSS OF REDUNDANT HARDWARE ITEMS COULD NOT RESULT FROM A SINGLE CREDIBLE EVENT.</p>	<p>1R HAZARD REF: ME-C1S,M ME-D1S,M</p>
SM 4-3	<p>FAILURE OF ONE QUALIFIED CHANNEL WITHIN REDLINE LIMITS RESULTS IN LOSS OF REDLINE PROTECTION. LOSS OF VEHICLE DUE TO TURBOPUMP FAILURE MAY RESULT IF EXCESSIVE VIBRATION OCCURS AND IS NOT DETECTED.</p> <p>REDUNDANCY SCREENS: CONTROLLER SYSTEM - TURBOPUMP SYSTEM: UNLIKE REDUNDANCY</p> <p>A: PASS. REDUNDANT HARDWARE ITEMS ARE CAPABLE OF CHECKOUT DURING NORMAL GROUND TURNAROUND. B: FAIL. LOSS OF A REDUNDANT HARDWARE ITEM IS NOT DETECTABLE DURING FLIGHT. C: PASS. LOSS OF REDUNDANT HARDWARE ITEMS COULD NOT RESULT FROM A SINGLE CREDIBLE EVENT.</p>	<p>1R HAZARD REF: ME-C1S,M ME-D1S,M</p>
SM 4-4	<p>FAILURE OF TWO REMAINING OR ALL QUALIFIED CHANNELS OUTSIDE OF REDLINE MONITOR LIMITS CAUSES ERRONEOUS ENGINE SHUTDOWN. MISSION ABORT.</p> <p>REDUNDANCY SCREENS: CONTROLLER SYSTEM - TURBOPUMP SYSTEM: UNLIKE REDUNDANCY</p> <p>A: PASS. REDUNDANT HARDWARE ITEMS ARE CAPABLE OF CHECKOUT DURING NORMAL GROUND TURNAROUND. B: PASS. LOSS OF A REDUNDANT HARDWARE ITEM IS DETECTABLE DURING FLIGHT. C: PASS. LOSS OF REDUNDANT HARDWARE ITEMS COULD NOT RESULT FROM A SINGLE CREDIBLE EVENT.</p>	<p>1R HAZARD REF: ME-C1S,M ME-D1S,M</p>

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CIL ITEM: F200-B2-01	DESIGN	DOCUMENT REF.
ALL CAUSES: INPUT ELECTRONICS: VIBRATION SIGNAL PROCESSOR ELECTRONICS (1)		(1) HONEYWELL BLOCK II FMEA VOL IV IE-F21
THREE VIBRATION SENSOR CHANNELS ARE PROVIDED FOR EACH HIGH PRESSURE TURBOPUMP (1). AGREEMENT BETWEEN 3 OF 3 CHANNELS (2 OF 3 IF THIRD CHANNEL IS DISQUALIFIED) CONSTITUTES A GOOD VOTE (2). A FAILURE MUST OCCUR IN TWO CHANNELS (MULTIPLE FAILURE) TO CAUSE LOSS OF FUNCTION (2). VIBRATION DIGITAL DATA IS TRANSMITTED IN THE VEHICLE DATA TABLE AND VIBRATION SHUTDOWN LIMIT EXCEEDED MONITORING CAN BE PRECLUDED BY OPERATIONAL DATA (2). THE CONTROLLER (WITH SOFTWARE) IS CONFIGURED TO DETECT AND RESPOND PROPERLY TO OUT OF QUALIFICATION LIMIT FAILURES, IMPLEMENT THE APPROPRIATE REDUNDANCY MANAGEMENT RESPONSE, AND COMMAND A SAFE ENGINE STATE WHEN CONTROLLER REDUNDANCY IS LOST (2). HOWEVER, THE CONTROLLER (WITH SOFTWARE) IS NOT CAPABLE OF DETECTING OR RESPONDING TO A FAILURE WHICH RESULTS IN THE LIMITS BEING MONITORED FAILING WITHIN ENGINE REDLINE LIMITS OR WITHIN NORMAL ENGINE OPERATING CONDITIONS (2).		(1) DSHG8977A1 (2) CP406R0008 3.1.3:3 3.2.1:6.2 3.2.3:2 3.2.3:3 3.2.3:4 3.2.3:5,4 3.2.4
ELECTRICAL, MECHANICAL, AND ELECTROMECHANICAL PARTS FOR THE CIRCUITS INVOLVED IN THIS FUNCTION HAVE BEEN SELECTED FROM THE CLASS B OR EQUIVALENT APPROVED PARTS SELECTION LIST (1), QUALITY ASSURANCE AND SCREENING REQUIREMENTS FOR HIGH RELIABILITY MICROCIRCUITS (2), AND THE SPECIFICATION CONTROL DRAWING FOR MICROCIRCUITS, MONOLITHIC SILEXON, AND TRANSISTOR-TRANSISTOR LOGIC (TTL) FAMILY OF DEVICES (3). PRIOR TO INSTALLATION ON THE CARD ASSEMBLIES, COMPONENTS ARE SUBJECTED TO A BURN-IN PERIOD WHERE PARTS ARE EXERCISED AT TEMPERATURES IN EXCESS OF NORMAL CONTROLLER OPERATING ENVIRONMENT, BUT LESS THAN COMPONENT MAXIMUM SPECIFIED OPERATING ENVIRONMENT, TO SCREEN FOR INFANT MORTALITY (4). IN ADDITION TO THESE REQUIREMENTS, ALL COMPONENTS ARE DERATED FROM THE MAXIMUM RATING AT OPERATING EXTREMES (5). CLEANLINESS AND ALL PROCESSES USED DURING MANUFACTURE ARE CONTROLLED BY SPECIFICATION REQUIREMENTS (5).		(1) B5M03928, B5M03929 (2) B5M02704 (3) B5M03766 (4) B5M03876 (5) DSHG8977A1
THE CONTROLLER DESIGN MEETS ALL CEI FAIL-OPERATE/FAILSAFE REQUIREMENTS FOR THIS FAILURE MODE (1). REDUNDANT CONTROLLER CHANNEL FUNCTIONS ARE PHYSICALLY SEPARATED WITHIN THE CONTROLLER HOUSING (2). CIRCUITS ARE DESIGNED TO PREVENT BRIDGING (2). ALL EEE PARTS ARE REQUIRED TO HAVE CONFORMAL COATING, AND INTERNAL WIRING IS INSULATED TO PREVENT SHORT CIRCUITS FROM CONDUCTIVE CONTAMINATION (3). ALL ELECTRICAL COMPONENTS ARE CONTAINED WITHIN THE CASE STRUCTURE WHICH IS PRESSURIZED WITH A POSITIVE PRESSURE INERT GAS BACKFILL TO PREVENT CONTAMINATION (2). WHERE APPLICABLE, ELECTROMAGNETIC INTERFERENCE SHIELDING IS PROVIDED AND CIRCUIT INTERCONNECTS USE TWISTED PAIR WIRING (2). ALL CIRCUIT CARDS ARE KEYS TO THEIR RESPECTIVE CONNECTION LOCATIONS TO PRECLUDE IMPROPER INSTALLATION (2). RAMP CLAMPS ARE UTILIZED TO PRECLUDE VIBRATION INDUCED CARD FAILURES (2). DESIGN OF CIRCUIT CARDS AND DETERMINATION OF COPPER PATH TRACE SPACING, WEIGHT, AND WIDTH IS CONTROLLED BY SPECIFICATION (2). EACH UNIT (PRODUCTION AND RECYCLE) IS REQUIRED TO PASS A FUNCTIONAL ACCEPTANCE TEST UNDER ENVIRONMENTAL CONDITIONS BEYOND THOSE SEEN DURING NORMAL FIELD OPERATION WITHOUT DEGRADATION OF HARDWARE LIFE EXPECTANCY (2). A WORST CASE CIRCUIT ANALYSIS WAS PERFORMED TO VERIFY NOMINAL OPERATION AT SPECIFICATION LIMITS (4). AN ANALYSIS WAS PERFORMED BY HONEYWELL TO ASSURE NO SINGLE POINT COMPONENT FAILURES ARE INHERENT TO THE CONTROLLER DESIGN (5).		(1) EP320R0003B (2) DSHG8977A1 (3) B5M03928 (4) ES24472-01 (5) HONEYWELL BLOCK II FMEA VOL I

CIL ITEM: F200-82-01		DESIGN	DOCUMENT REF.
		DESIGN QUALIFICATION TESTING OF THE CONTROLLER ASSEMBLY HAS BEEN PERFORMED [INCLUDING: FAULT INSERTION TESTING (1), ASSEMBLY THERMAL CYCLING (2), CASE AND ASSEMBLY VIBRATION TESTING (3), ELECTROMAGNETIC INTERFERENCE AND SUSCEPTIBILITY TESTING (4), ACOUSTICAL NOISE TESTING (5), CASE ULTIMATE PRESSURE TESTING (6), AND ASSEMBLY BREAK OPEN INSPECTION (7)].	(1) TR34079282 TR34080202 TR34080258 TR34080259 (2) TR34080203 TR34080207 (3) TR34080204 TR34080205 TR34080206 TR34087499 (4) TR34080209 (5) TR34080210 (6) TR34085021 (7) TR34085022
CIL ITEM: F200-82-01		INSPECTION AND TEST	
POSSIBLE CAUSES	SIGNIFICANT CHARACTERISTICS	INSPECTION(S)/TEST(S)	DOCUMENT REF.
ALL CAUSES:	RE1493 - CONTROLLER		RE1493
	ASSEMBLY INTEGRITY	CLEANLINESS REQUIREMENTS ARE VERIFIED PER SPECIFICATION DURING MANUFACTURING OF THE CONTROLLER ASSEMBLY. ENVIRONMENT CONTROLS (TEMPERATURE, HUMIDITY) ARE ENFORCED DURING ASSEMBLY AND TESTING PER SPECIFICATION REQUIREMENTS. TO PREVENT COMPONENT DAMAGE, STATIC ELECTRICAL DISCHARGE POTENTIAL IS CONTROLLED DURING MANUFACTURING PER SPECIFICATION REQUIREMENTS.	RC1493 DSHG8977A1 DSHG8977A1
	MATERIAL INTEGRITY	MATERIAL INTEGRITY IS VERIFIED PER SPECIFICATION REQUIREMENTS.	RC1493

CIL ITEM: F200-BZ-01		INSPECTION AND TEST	
POSSIBLE CAUSES	SIGNIFICANT CHARACTERISTICS	INSPECTION(S)/TEST(S)	DOCUMENT REF.
		PROTECTIVE FINISHES AND MATERIAL SELECTION TO PREVENT DETRIMENTAL EFFECTS FROM ENVIRONMENTAL EXPOSURE, STRESS CORROSION, AND ELECTROLYTIC CORROSION ARE VERIFIED PER SPECIFICATION REQUIREMENTS.	RC1493 MSFC-SPEC-250
		FLAMMABILITY REQUIREMENTS ARE VERIFIED PER SPECIFICATION REQUIREMENTS.	RC1493
	INTEGRITY OF ELECTRONICS	THE FOLLOWING PROCESSES ARE VERIFIED PER SPECIFICATION DURING MANUFACTURING OF THE CARD ASSEMBLIES: <ul style="list-style-type: none"> - CONSTRUCTION OF PRINTED CIRCUIT BOARDS. - INSTALLATION OF TERMINALS. - PLASTICS AND ELASTOMERS FOR ELECTRONIC ENCAPSULATION. - SOLDERED ELECTRICAL CONNECTIONS. - POST-SOLDERING INSPECTION FREE OF SPLATTER AND CONTAMINATION. - ELECTRICAL BONDING. - COMPONENT LEAD AND INTERCONNECTION MATERIAL SELECTION. - FREE OF CONTAMINATION AFTER CONFORMAL COATING. 	RL10005 RL10007 RL10008 RL10009 RL10009 RC1493 BSM03928 RC1493
	WELD INTEGRITY	ALL WELDS ARE VERIFIED TO DRAWING AND SPECIFICATION REQUIREMENTS.	RL10011
	PRE-CLOSEOUT TESTING AND INSPECTION	THE FOLLOWING TESTS ARE PERFORMED AT THE CARD/COMPONENT LEVEL DURING MANUFACTURING: <ul style="list-style-type: none"> - PARTIAL IMPACT NOISE DETECTION (PIND) TEST ON HYBRID MICROCIRCUITS AND CAVITY TYPE DEVICES. - ULTRASONIC SCAN TEST FOR DELAMINATION CERAMIC ON CAPACITORS (CKR05 AND CKR06). - BURN-IN PERIOD FOR ELECTRICAL PARTS. - INSULATION RESISTANCE AND CONTINUITY TEST. - DIELECTRIC WITHSTANDING VOLTAGE. - FUNCTIONAL TEST. - X-RAY OF CONTROLLER INTERNAL CABLES AND WIRES. 	DSHG8977A1 DSHG8977A1 BSM03928 OSHG8977A1 DSHG8977A1 DSHG8977A1 DSHG8977A1

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CIL ITEM: F200-B2-01		INSPECTION AND TEST	
POSSIBLE CAUSES	SIGNIFICANT CHARACTERISTICS	INSPECTION(S)/TEST(S)	DOCUMENT REF.
		<p>THE FOLLOWING INSPECTIONS ARE PERFORMED ON THE COMPLETED ASSEMBLY PRIOR TO FINAL CLOSURE:</p> <ul style="list-style-type: none"> - INTERNAL AND EXTERNAL CLEANLINESS. - CARD DIMENSIONS ARE VERIFIED AT CARD ASSEMBLY. - CHASSIS CAVITY INSPECTION FOR CONTAMINATION. - ELECTRICAL INTERFACE CONNECTORS. - CARD INSTALLATION AND CONFIGURATION AUDIT. - CONFORMAL COATING OF EEE PARTS. 	RC1493 DSHG8977A1 RC1493 RC1493 DSHG8977A1 B5N03928
	ACCEPTANCE TESTS	<p>THE FOLLOWING TESTS ARE PERFORMED BY HONEYWELL DURING ACCEPTANCE TESTING:</p> <ul style="list-style-type: none"> - HERMETIC SEAL AND PRESSURIZATION PORT LEAK TEST. - FUNCTIONAL TEST INCLUDING: <ul style="list-style-type: none"> - OUTPUT INTERFACE, - CHECKOUT, - OPERATION, - CONTROLLER CHECKOUT. - HIGH TEMPERATURE OPERATION. - LOW TEMPERATURE OPERATION. - VIBRATION TESTING. - FINAL FUNCTIONAL TEST. 	RC1493 RC1493 RC1493 RC1493 RC1493 RC1493 RC1493 RC1493 RC1493
	HOT FIRE ACCEPTANCE TESTING (GREEN RUN)	CONTROLLER OPERATION IS VERIFIED THROUGH ENGINE HOT FIRE ACCEPTANCE TESTING.	RL00461
	HERMETIC SEAL INTEGRITY	INTERNAL PRESSURE IS MONITORED DURING CONTROLLER OPERATION.	OMRSD V41AWO.040
	FLIGHT FLOW TESTING	<p>THE FOLLOWING TESTS ARE PERFORMED DURING FLIGHT FLOW VEHICLE PROCESSING AND AFTER ANY MAINTENANCE OR REPLACEMENT:</p> <ul style="list-style-type: none"> - CONTROLLER CHECKOUT. - SENSOR CHECKOUT. - SELF-TEST. 	OMRSD V41AWO.035 OMRSD V41AWO.010 CP406R0008

CIL ITEM: F200-B2-01		INSPECTION AND TEST	
POSSIBLE CAUSES	SIGNIFICANT CHARACTERISTICS	INSPECTION(S)/TEST(S)	DOCUMENT REF.
	PRE-FLIGHT CHECKOUT	THE FOLLOWING TESTS ARE PERFORMED DURING LAUNCH COUNTDOWN: - PRE-CRYO LOADING CHECKOUT INCLUDING CONTROLLER, SENSORS, AND PNEUMATICS CHECKOUT. - SELF-TEST.	ONRSD 800FAD.213 CP406R008
FAILURE HISTORY: COMPREHENSIVE FAILURE HISTORY DATA IS MAINTAINED IN THE PROBLEM REPORTING DATABASE (PRMS/PRCA). REFERENCE: NASA LETTER 8421/88/308 AND ROCKETDYNE LETTER 88RCD9761.			

OPERATIONAL USE: NOT APPLICABLE.